

Unsteady pulsating flowfield over spiked axisymmetric forebody at hypersonic flows

Cite as: Phys. Fluids **34**, 016104 (2022); <https://doi.org/10.1063/5.0075583>

Submitted: 16 October 2021 • Accepted: 17 December 2021 • Published Online: 05 January 2022

 Mohammed Ibrahim Sugarno,  R. Sriram,  S. K. Karthick, et al.



View Online



Export Citation



CrossMark

ARTICLES YOU MAY BE INTERESTED IN

[Shock and shear layer interactions in a confined supersonic cavity flow](#)
Physics of Fluids **33**, 066102 (2021); <https://doi.org/10.1063/5.0050822>

[Low-frequency shock train oscillation control in a constant area duct](#)
Physics of Fluids **34**, 016105 (2022); <https://doi.org/10.1063/5.0071500>

[On the unsteady throttling dynamics and scaling analysis in a typical hypersonic inlet-isolator flow](#)
Physics of Fluids **32**, 126104 (2020); <https://doi.org/10.1063/5.0032740>

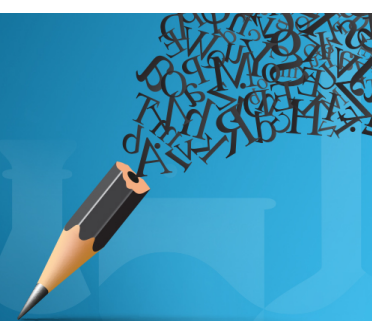


Author Services

English Language Editing

High-quality assistance from subject specialists

LEARN MORE



Unsteady pulsating flowfield over spiked axisymmetric forebody at hypersonic flows

Cite as: Phys. Fluids **34**, 016104 (2022); doi: [10.1063/5.0075583](https://doi.org/10.1063/5.0075583)

Submitted: 16 October 2021 · Accepted: 17 December 2021 ·

Published Online: 5 January 2022



View Online



Export Citation



CrossMark

Mohammed Ibrahim Sugarno,^{1,a)}  R. Sriram,²  S. K. Karthick,³  and Copalan Jagadeesh⁴ 

AFFILIATIONS

¹Department of Aerospace Engineering, Indian Institute of Technology, Kanpur 208016, India

²Department of Aerospace Engineering, Indian Institute of Technology-Madras, Chennai 600036, India

³Faculty of Aerospace Engineering, Technion-Israel Institute of Technology, Haifa 3200003, Israel

⁴Department of Aerospace Engineering, Indian Institute of Science, Bengaluru 560012, India

^{a)} Author to whom correspondence should be addressed: ibrahim@iitk.ac.in

ABSTRACT

The paper gives experimental observations of the hypersonic flow past an axisymmetric flat-face cylinder with a protruding sharp-tip spike. Unsteady pressure measurements and high-speed schlieren images are performed in tandem on a hypersonic Ludwig tunnel at a freestream Mach number of $M_\infty = 8.16$ at two different freestream Reynolds numbers based on the base body diameter ($Re_D = 0.76 \times 10^6$ and 3.05×10^6). The obtained high-speed images are subjected further to modal analysis to understand the flow dynamics parallel to the unsteady pressure measurements. The protruding spike of length to base body diameter ratio of $[l/D] = 1$ creates a familiar form of an unsteady flowfield called “pulsation.” Pressure loading and fluctuation intensity at two different Re_D cases are calculated. A maximum drop of 98.24% in the pressure loading and fluctuation intensity is observed between the high and low Re_D cases. Due to the low-density field at low Re_D case, almost all image analyses are done with the high Re_D case. Based on the analysis, a difference in the pulsation characteristics is noticed, which arises from two vortical zones, each from a system of two “ λ ” shocks formed during the “collapse” phase ahead of the base body. The interaction of shedding vortices from the λ -shocks’ triple-points, along with the rotating stationary waves, contributes to the asymmetric high-pressure loading and the observation of shock pulsation on the flat-face cylinder. The vortical interactions forming the second dominant spatial mode with a temporal mode carry a dimensionless frequency ($f_2 D/u_\infty \approx 0.34$) almost twice that of the fundamental frequency ($f_1 D/u_\infty \approx 0.17$). The observed frequencies are invariant irrespective of the Re_D cases. However, for the high-frequency range, the spectral pressure decay is observed to follow an inverse and $-7/3$ law for the low and high Re_D cases, respectively.

Published under an exclusive license by AIP Publishing. <https://doi.org/10.1063/5.0075583>

I. INTRODUCTION

Shock-shock^{1–4} and shock-boundary-layer^{5–7} interaction is both complex and an interesting flowfield to study in the domain of supersonic/hypersonic aerodynamic research. In the past, several kinds of research^{8–11} have been done to understand the nature of the flowfield, interacting shocks influences and methods to control them.^{3,12–14} An important aspect of such a flowfield is the unsteadiness induced by the interaction,^{15–19} which can either be local or global, as in the case of self-sustained oscillatory flows.^{20–23} The effects of local unsteadiness on the flowfield are minimal and can be mitigated using control techniques. However, the global unsteadiness specifically induced by the geometrical configuration significantly modifies the overall flowfield.^{24,25} One form of global unsteadiness is the flowfield over spiked forebodies.^{26,27}

Spiked forebodies were found to be very effective in reducing the aerodynamic drag in high-speed vehicles^{24,27} among the other techniques. Rockets and missiles traveling at supersonic/hypersonic speeds within the atmosphere are subjected to severe aerodynamic heating and drag forces.^{28,29} The problem of heating is minimized by having a blunt forebody; however, it comes with a penalty of increased drag.^{30,31} Such blunt forebody shapes are only preferred for atmospheric re-entry vehicles where minimizing aerodynamic heating is a priority.³² However, in the case of vehicles traveling at high speeds within the atmosphere, reducing aerodynamic drag is of utmost importance to enhance its range and efficiency. The spiked forebody in any form, either in missiles^{33,34} or engine intakes,^{35,36} reduces the overall drag as the spike creates a low pressure, recirculating, dead air region in front of the forebody. The pressure forces acting on the

forebody are relatively lower than those acting on a forebody with no spike, resulting in drag reduction.

However, in spiked forebodies for a certain range of $[l/D]$ ratios^{37–39} (ratio of spike’s length to base body diameter), the flowfield is either pulsating⁴⁰ ($l/D < 1.4$) or oscillating⁴¹ ($1.4 < [l/D] < 2.5$). Pulsation is characterized by sudden collapse and rapid expansion of the forebody shock and separation region. In oscillation, the forebody shock changes its shape from concave to convex. Both these flow modes are highly unsteady, cyclic in nature, and geometry dependent, which has motivated many researchers to study them beginning in the early 1950s.^{24,27,42,43} A detailed list of studies on spiked body flows for a wide range of freestream Reynolds numbers based on the base body diameter (Re_D) and freestream Mach numbers is shown in Fig. 1. Most of them are experiments, and a few of them are computations. The plot is generated from the collective literature available on the spiked body flow physics presented in Table 1 of Sahoo *et al.*⁴⁴ Surprisingly, from the figure, it can be seen that there are scarcely a few cases available at high M_∞ and high Re_D .

As mentioned earlier, the unsteady pulsation flow modes have been studied by researchers in the past both experimentally and numerically.^{45–49} Several theories have been proposed concerning the driving mechanism for these flows.^{24,27,50,51} Feszty *et al.*⁴⁰ carried out a numerical investigation, studying the laminar flowfield and the driving mechanism for pulsation over a spiked cylinder configuration with $[l/D] = 1$. They identified the following processes to occur in a pulsation cycle, namely collapse, inflation, and withhold, based on which

the driving mechanism was explained in detail. A vortical region is formed near the cylinder-spike junction during the collapse stage. It is the high-pressure gas that was trapped in this vortical region expanding violently during the inflation stage rather than the continuous mass influx due to Edney’s type-IV interaction.⁵² These features are identified as the driving mechanism for the self-sustained shock motion called pulsation, unlike the other means that were previously thought by other researchers.^{24,27,42,43} Even some of the results from the laminar flow numerical investigations were found to agree with the experimental work (flowfield visualization) of Kenworthy.³⁹

However, our recent experimental campaign on pulsating flowfields, especially over a spiked-blunt-forebody of base diameter D at high Re_D and freestream M_∞ , revealed exciting results, especially during the collapse phase of pulsation, which made us re-think the driving mechanism behind pulsation. The motivations behind our current experimental campaign are twofolds: (1) at high Re_D , the turbulent flow effects might be strong, which is different from the laminar observation done by Feszty *et al.*⁴⁰ In fact, from the brief literature review, the authors found that the research works at high Re_D and high M_∞ on spiked body flows are scarcely available (see Fig. 1); (2) high-speed schlieren imaging and pressure measurements at hypersonic Mach numbers will shed valuable information on the formation of typical flow structures that reveal the alterations in the pressure loading on the vehicle itself. If a clear understanding of the flowfield mentioned above is attempted, then only formulating an efficient active or passive control device would be feasible.

With the motivations mentioned above, the authors took the following as their distinct objectives for the present work:

1. To experimentally study the pulsating flowfield over a cylindrical forebody with a spike of $[l/D] = 1$, in a hypersonic flow with $M_\infty = 8.1$ at two different Re_D .
2. To obtain the pressure distribution on the flat-face cylindrical forebody and to visualize the flowfield using high-speed Schlieren imaging.
3. To compute and compare the variations in the pressure loading, fluctuation intensity, and spectral decay from the unsteady pressure measurements.
4. To understand the driving flow modes from the high-speed schlieren images after subjecting them through modal analysis (Proper Orthogonal Decomposition-POD and Dynamic Mode Decomposition-DMD).

The rest of the paper is organized as follows: details about the experimental methodology is given in Sec. II followed by the experimental uncertainties in Sec. III. Results and discussions are given in Sec. IV under different subsections: high-speed schlieren images in Sec. IV A, $x-t$ diagram in Sec. IV B, unsteady pressure signals in Sec. IV C, and the modal analysis in Sec. IV D. Some of the major conclusions are presented in Sec. V.

II. EXPERIMENTATION

A. Test facility

Experiments are performed in the recently developed IISc Ludwig tunnel, and the schematic (not drawn to scale) is shown in Fig. 2(a). Similarly, details about the testing model are given in Fig. 2(b). More details about the test model are given in Sec. II B. The Ludwig tunnel is a modification of the existing hypersonic shock

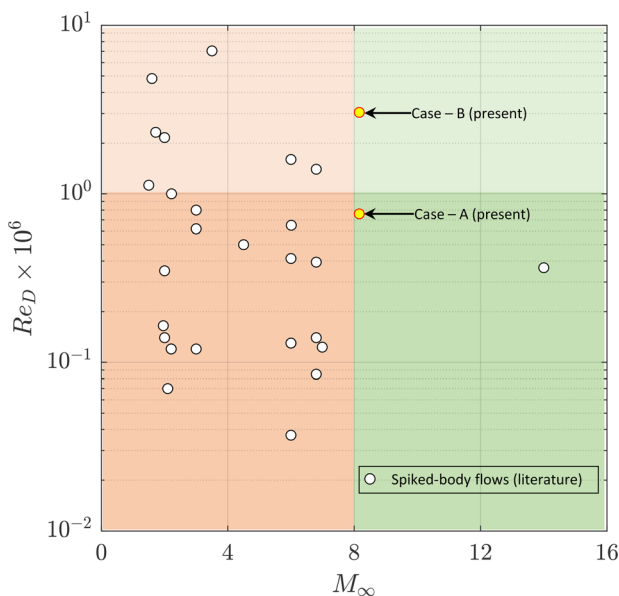


FIG. 1. Graphical representation showing the list of spiked-body flow experiments performed over the last millennia as reported in the work of Sahoo *et al.*⁴⁴ in the scatterplot format. The solid white-colored circles (black-outlined) show the freestream Mach number (M_∞) vs freestream Reynolds number based on the base body diameter (Re_D) variations. The present experimental cases are marked as solid yellow-colored circles (red-outlined). Orange and green shades demarcate the low and high M_∞ regime, whereas the light and dark shades represent the corresponding zones of the low and high Re_D regime.

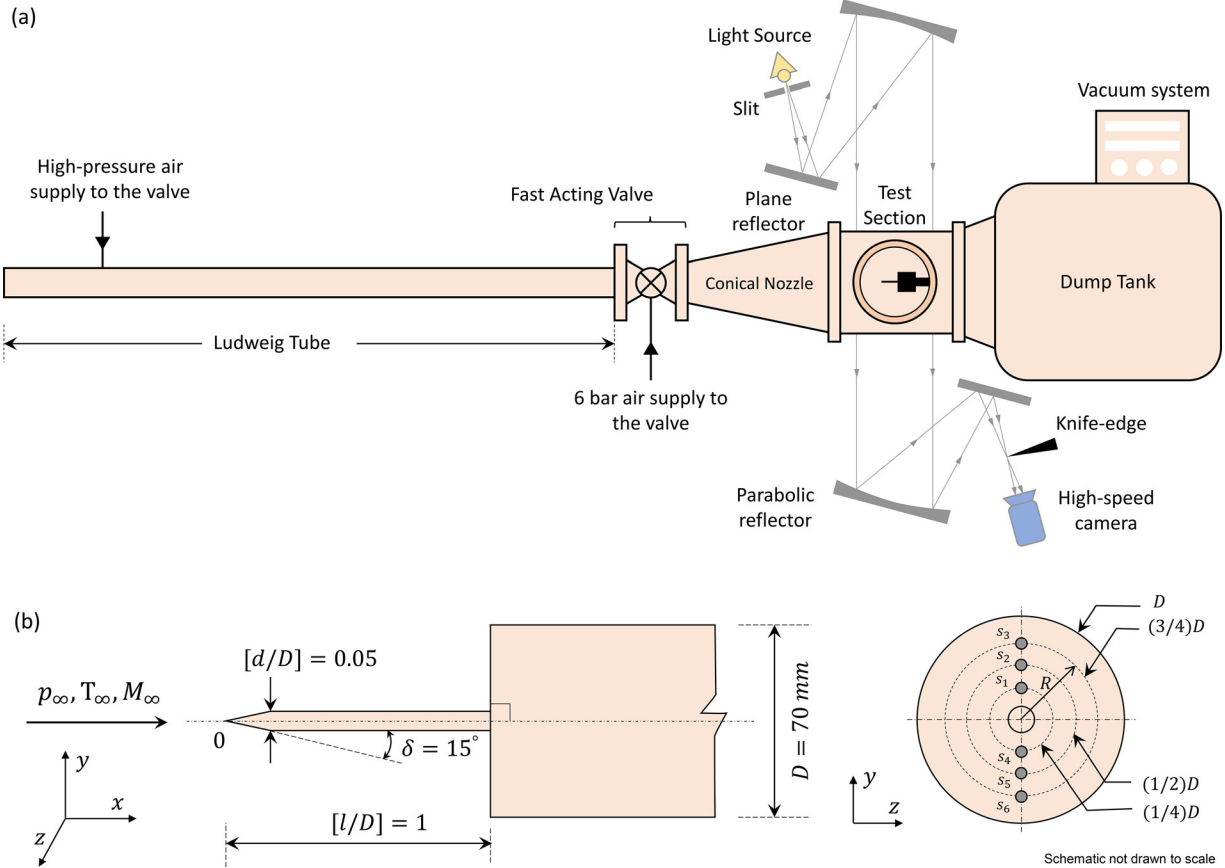


FIG. 2. (a) A schematic (not drawn to scale) showing the top view of the Ludweig facility at IISc-Bengaluru with the Z-type schlieren arrangement⁵³ to study the pulsating flow-field observed around an axisymmetric flat-face spiked body in a hypersonic flowfield; (b) a schematic (not drawn to scale) showing the geometrical configuration of the model under investigation and the locations of the unsteady pressure transducers ($S_1 - S_6$) mounted at different radial locals (R/D). Model is oriented against the flow direction.

tunnel: HST-2,⁵⁴ by merging the driver and driven sections to form a Ludweig tube. It is essentially a pressure tube where the test gas (nitrogen, N_2) is filled to the required pressure (p_{01}). The end of the Ludweig tube (length 7.12 m and inner diameter 50 mm) is connected to a commercially available fast-acting valve, ISTA[®] KB-40. It is a pneumatically assisted solenoid valve, requiring 6 bar pressure and 24 V DC for its operation. The valve response time for the input volt pressure is around 1 ms. The valve isolates the dump tank, containing the nozzle and test section at an ultra-low vacuum of 10^{-8} bar.

The operation of the valve results in the expansion of high-pressure gas in the pressure tube into the test section of cross section $300 \times 300 \text{ mm}^2$ and a length of 450 mm while passing through the diverging nozzle of design Mach number $M_{\infty,D} = 8$ [see Fig. 2(a)]. Later, the flow exits into the 2.5 m long dump tank. The facility is designed to generate the desired freestream flow with a uniform core flow diameter of 240 mm and unit Reynolds number varying from $10\text{--}90 \times 10^6/\text{m}$ depending on the fill pressure. The present experiments consider two different fill pressures to simulate two different Re_D : (a) 10 bar for low a Re_D of 0.76×10^6 and (b) 40 bar for a Re_D of 3.01×10^6 . The respective Re_D are selected in a manner to simulate

the flight conditions at altitudes of 22 and 13 km, where the spiked body flows have relevance in terms of axisymmetric scramjet inlet operation^{55,56} and ballistic missile drag reduction.^{24,26,27,57}

The Pitot pressure (p_{02}) is measured simultaneously along with the test model during the experiments in order to ascertain the freestream conditions (see Table 1). The achieved test time during a typical high Re_D case is shown in Fig. 3(a). A steady flow test period of around 25 ms was observed for the reported experiment, as seen from the Pitot pressure signal. In the present set of experiments, the facility was operated to give two different Reynolds numbers of 11×10^6 and $43 \times 10^6/\text{m}$, by filling the Ludwig tube with nitrogen at $p_{01} = 10$ and $p_{01} = 40$ bar, respectively. A typical normalized pressure amplitude spectral decay observed from the s_4 unsteady pressure sensor is plotted in Fig. 3(b) for two different Re_D cases. The dimensionless frequency is given by $[fD/u_\infty]$, where the freestream velocity as tabulated in Table 1 remains constant for both the cases. Initial observation shows that the dominant components remain invariant with Re_D . More details of the unsteady pressure spectra are briefly discussed in the upcoming sections, particularly at Sec. IV C.

TABLE I. Freestream flow conditions achieved in the test section of the Ludwig tunnel during the present investigation for two different freestream Reynolds number (Re_D) cases based on the base body diameter (D) at a constant freestream Mach number of $M_\infty = 8.16$.

Quantities	Case-A (low Re_D)	Case-B (high Re_D)
Total pressure (p_{01} , Pa)	$10 \times 10^5 \pm 5\%$	$40 \times 10^5 \pm 5\%$
Total temperature (T_{01} , K)	$300\% \pm 2\%$	$300\% \pm 2\%$
Freestream pressure (p_∞ , Pa)	$90.05\% \pm 5\%$	$360.19\% \pm 5\%$
Freestream temperature (T_∞ , K)	$20.95\% \pm 2\%$	$20.95\% \pm 2\%$
Freestream density (ρ_∞ , kg/m ³)	$0.02\% \pm 5\%$	$0.06\% \pm 5\%$
Freestream velocity (u_∞ , m/s)	$748.66\% \pm 2\%$	$748.66\% \pm 2\%$
Freestream kinematic viscosity (ν_∞ , m ² /s)	$6.87 \times 10^{-5} \pm 2\%$	$1.72 \times 10^{-5} \pm 2\%$
Freestream Mach number (M_∞)	$8.16\% \pm 1\%$	$8.16\% \pm 1\%$
Reynolds number based on $D = 70$ mm ($Re_D = u_\infty D / \nu_\infty$)	$0.76 \times 10^6 \pm 5\%$	$3.05 \times 10^6 \pm 5\%$
Equivalent altitude (h , km)	21.8	13.2

B. Test model, instrumentation, and flow diagnostics

The test model used is a flat-face cylinder with a diameter of $D = 70$ mm, having a forward-facing spike of length $l = 70$ mm ($l/D = 1$, fineness ratio) with a conical tip having a semi-apex angle of $\delta = 15^\circ$. The spike has a slenderness ratio of $[d/D] = 0.05$, where d is the diameter of the spike. The model is instrumented with high-frequency pressure transducers (PCB Piezotronics) of 1 MHz acquisition rate flush-mounted on its flat-face at radial (R) distances of 17.5 ($D/4$), 35 ($D/2$), and 52.5 ($3D/4$) mm from the axis along the transverse direction, to measure the unsteady pressure fluctuations. A schematic of the test model and the sensor mounting location is shown in Fig. 2(b).

The hypersonic pulsating flowfield over the test model was visualized using the “Z-type” high-speed schlieren technique⁵³ using a Phantom V310 high-speed camera, operated at 40 kHz, with an

exposure time of $2 \mu\text{s}$. The frame size was 256×256 pixels with a pixel resolution of ~ 0.45 mm/pixel. A typical instantaneous schlieren image showing the normalized line-of-sight integrated density gradients in y -direction is shown in Fig. 4 (Multimedia view) for two different Re_D cases. The sensitivity of schlieren imaging plays a vital role in resolving the flow features. Therefore, flow features are not captured with good contrast in Fig. 4(a) owing to the low density, but they are in Fig. 4(b). The schlieren setup, as shown in Fig. 2(a), utilizes a light source made from an array of white LEDs of 7 W power after passing through a pinhole slit. The knife-edge was kept horizontal at the point of focus, enabling us to see the light intensity changes due to density gradients in the vertical direction ($\partial\rho/\partial y$), as the flow features [shock–shock interaction, separation region, and vortical regions, see Fig. 5 (Multimedia view)] were visualized only in this orientation of the knife-edge.

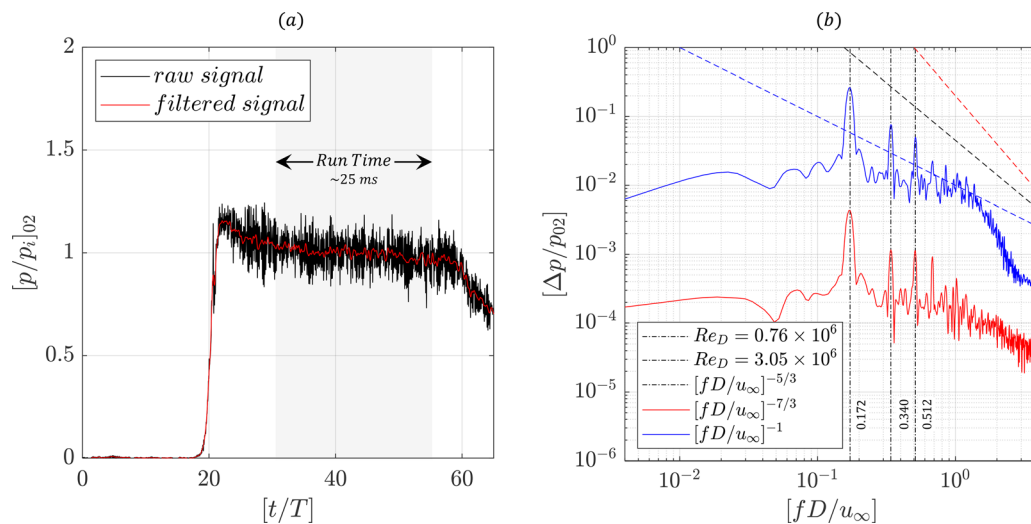


FIG. 3. (a) A typical pitot pressure signal (p_{02}) shows a region of almost constant run-time from the mounted pitot tube in the Ludwig tunnel’s test section. The x-axis is normalized with $T = 1$ ms and the y-axis is normalized with the ideal or isentropic pitot-pressure ($p_{02,i}$) based on the design Mach number ($M_{\infty,D}$); (b) normalized pressure amplitude spectra ($\Delta p/p_{02}$) showing the dimensionless spectral decay (fD/u_∞) observed from the unsteady pressure signal at s_4 sensor location for two different Re_D cases. The decay rate (dash lines) is also compared with the inverse law, $-7/3$ law, and $-5/3$ law for evaluating the behavior of turbulent structures formed during the pulsation events.

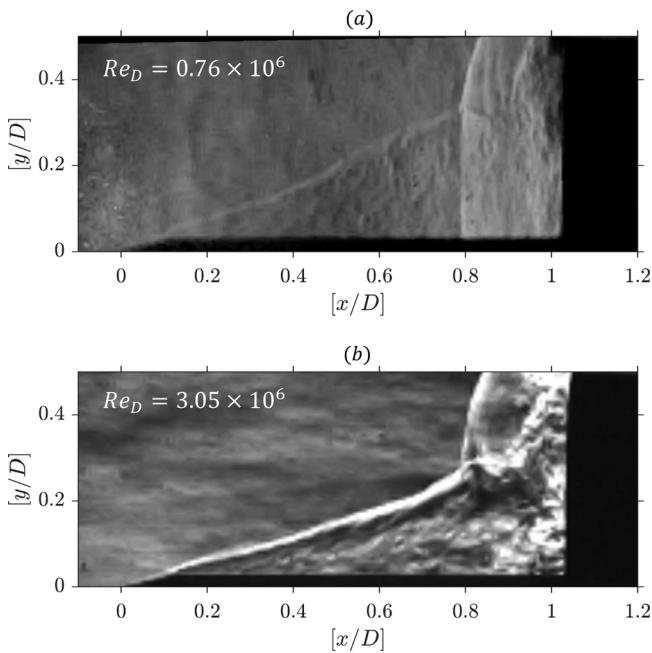


FIG. 4. A typical instantaneous schlieren image depicting the line-of-sight integrated normalized density gradients in y -direction $[|(\partial\rho/\partial y)|]$ taken at an arbitrary time interval (τ) for two different freestream Reynolds numbers based on the base body diameter (Re_D) at a freestream Mach number of $M_\infty = 8.16$: (a) $Re_D = 0.76 \times 10^6$ (low Re_D , case-A), and (b). $Re_D = 3.01 \times 10^6$ (high Re_D , case-B). Detailed flow features are marked in Fig. 5 as they are almost common for both cases. Multimedia view: <https://doi.org/10.1063/5.0075583.1>

III. MEASUREMENT UNCERTAINTIES

All the experiments were repeated at least three times to ensure statistical consistency and repeatability. Steady and unsteady

measurements suffer from uncertainty due to repeatability, acquisition, data conversion, storage, and sensitivity to external factors. The steps are given in the text of Coleman and Steele⁵⁸ and the recommendations while taking pitot measurements in Sutcliffe, and Morgan⁵⁹ were followed to compute the total uncertainty in pressure data. Uncertainties from the images and the derived data were computed using the principles given by Santo *et al.*⁶⁰ All the signals were preconditioned, and no further post-processing was done, including padding, windowing, or spectral smoothing. Measurements of unsteady pressure and schlieren imaging were done simultaneously to resolve and associate the flow features with the respective dynamic events. All the steady-state or low-response pressure transducers like the driver pressure monitor or p_{01} include a total uncertainty of $\pm 5\%$ about the mean. The unsteady pressure transducers also exhibit a total uncertainty of $\pm 5\%$ about the mean with a spectral resolution of $\Delta f_p \approx 40$ Hz [the total number of considered samples (n) is 25 001, and the sampling rate (f_s) is 1 MHz which results in the spectral resolution of $\Delta f = f_s/n = 39.9984$ Hz]. The schlieren images and the resulting spatial modes from the modal decomposition exhibit a spatial resolution of $\Delta x \approx \Delta y \approx 0.5$ mm. The spectral resolution from the modal decomposition is $\Delta f_m \approx 40$ Hz.

IV. RESULTS AND DISCUSSION

A. Flow physics from high-speed schlieren images

A pulsating flowfield around the spiked body has many flow features. High-speed schlieren helps in identifying them. A typical instantaneous and time-averaged schlieren image is given in Fig. 5. As the low Re_D case results in low density and the sensitivity of the schlieren imaging system has limitations in resolving small density gradients, hereafter in the discussions, only the high Re_D case is considered unless otherwise specified. Some of the flow features, including separated free shear layer, vortices from the triple point, and recirculation region, can be seen from the instantaneous image [Fig. 5(a)]. The vortical structures are identified with surety because previously, in Feszty *et al.*,⁴⁰

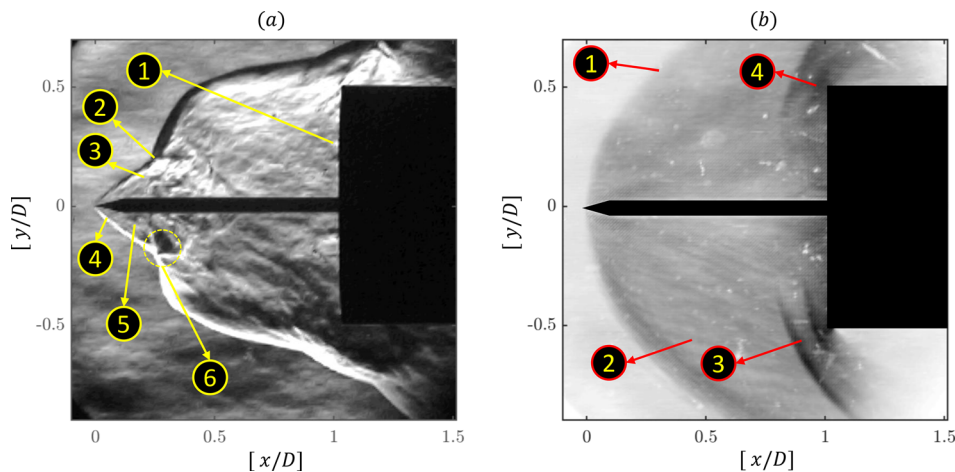


FIG. 5. (a) Instantaneous schlieren imaging showing the pulsating flowfield at an instant of the collapse phase for case-B (high Re_D); dominant flow features: (1) gas compression due to collapse in front of the flat-face, (2) triple point (λ -shock), (3) separated free shear layer, (4) leading-edge shock, (5) recirculation zone, (6) shedding vortices from the triple point. (b) Operator based time-averaged image ($||\bar{I} - I_{rms}||$) showing the extent of pulsating shock-laden flowfield. Dominant flow features: (1) inflated shock in the leading edge of the spike; (2) trace of the separation shock during the time of inflation; (3) collapsed shock forming ahead of the flat-face; (4) shock formation due to the rapid compression of collapsing flowfield. Multimedia view: <https://doi.org/10.1063/5.0075583.2>

the authors matched the vortical zones from their computations with that of the schlieren images taken by Kenworthy.³⁹ The comparison further revealed the locations where the vortical zones are expected in the schlieren images.

One of the dominant flow features like the moving shocks from the flat-face to the spike-tips captured in Fig. 5(b) where the operator-based time-averaged image ($|\bar{I} - I_{rms}|$) is shown. The presence of two strong shocks in front of the forebody is vital to note, as it is the characteristics observed at higher Re_D and M_∞ , which was not observed before. In addition, the shock angle from the leading edge of the spike-tip is shallower at higher freestream hypersonic Mach numbers than that of the lower supersonic freestream Mach numbers. The shallow shock angle meets the forebody halfway and impinges on the solid forebody resulting in the formation of Edney's type-IV⁵² shock interference pattern. The associated flow events have similarities with relevant pulsating flowfields at other speed regimes; however, they are not identical.^{24,27,39,40} Respective discussions toward those non-identical flow events are explained further.

A pulsation cycle consists of three stages. All three stages are shown in Fig. 6 (Multimedia view) as sequence of images, starting from $\tau = 0$ to $\tau + 19\Delta\tau = 475 \mu\text{s}$ with $\Delta\tau = 25 \mu\text{s}$. The start to end of a pulsation cycle (frames Ⓐ–Ⓜ) and the events corresponding to the observations are also marked on the obtained pressure signals, near root ($s_1 = 0.25D$) and shoulder ($s_3 = 0.75D$) region of the test model, as shown in Figs. 7(a) and 7(b).

Collapse stage (marked between frame Ⓐ at τ to frame Ⓜ at $\tau + 7\Delta\tau$) is characterized by unsteady shock motion from the tip of the spike toward the after body. "Inflation" stage (marked between frame Ⓜ at $\tau + 8\Delta\tau$ to frame Ⓐ at $\tau + 15\Delta\tau$) follows next, where the high-pressure gas trapped during the collapse stage expands rapidly in the recirculation region. "With-hold" (marked between frame Ⓐ at $\tau + 16\Delta\tau$ to frame Ⓜ at $\tau + 19\Delta\tau$) is the last stage where the shock remains stationary and the exploding high pressure gas escapes through the shoulder of the cylindrical after body and initiates a collapse cycle.⁴⁰

Our discussion focuses mainly on the collapse stage of a pulsation cycle. During this stage, the unsteady shock-shock interaction has led to the formation and merging of toroidal vortices ($\tau + 9\Delta\tau$), rapidly growing in size and contributing to a sustaining event of pulsation.

The cycle starts with the interaction of oblique shock emanating from the tip of the spike with the bow body shock wave as seen in Ⓐ at τ . The shock system described above, named Converging Shock System (CSS), is unsteady and starts moving toward the right, as can be seen in the subsequent frames (till frame Ⓐ at $\tau + 15\Delta\tau$). The shock interaction, mainly seen in frame Ⓜ at $\tau + 7\Delta\tau$, is similar to the pattern described by Edney.⁵² However, it is unclear whether it is a type-III or type-IV interaction⁵² from the present flow visualization. At the point of interaction (from frame Ⓜ at $\tau + 7\Delta\tau$ to frame Ⓜ at $\tau + 9\Delta\tau$), i.e., triple point, a shear layer exists, due to the velocity gradient,^{61,62} forming the boundary between the subsonic flow behind the bow shock and supersonic flow behind the reflected oblique shock wave. The reflected oblique shock wave impinges on the boundary layer of the spike's body surface, causing flow separation, which shall be discussed in the subsequent paragraph.

At the same time, there is a rapid expansion of high-pressure gas at supersonic speeds,⁶³ in the opposite direction to the motion of the shock system, corresponding to the inflation stage of the previous

pulsation cycle (supersonic jet with shock-cells in frame Ⓐ at τ). At the time of collapse, there is a clear demarcation between the two flow-fields as can be very clearly seen in frame Ⓜ at $\tau + 2\Delta\tau$ in the form of a beetle leaf-like structure. This demarcation line/boundary is a shock wave separating the supersonic flow on the side (cylindrical afterbody side) to the mixed subsonic/supersonic flow on the other side (spike tip side). With time, the beetle leaf-like structure expands in size (from Ⓜ at $\tau + 3\Delta\tau$ to Ⓜ at $\tau + 5\Delta\tau$). The beetle leaf-like structure then runs along the spike and then impinges on the cylindrical afterbody as seen in frame Ⓜ at $\tau + 6\Delta\tau$ and moves laterally along its surface, as seen in subsequent frames Ⓜ at $\tau + 6\Delta\tau$ and Ⓜ at $\tau + 7\Delta\tau$.

The shock wave impinges on the forebody causing peak pressure loads on the forebody (frame Ⓜ at $\tau + 11\Delta\tau$). The shock wave then moves over the shoulder, exposing the aft body region near the root of the spike to supersonic flow. The phenomenon mentioned above explains the formation of a normal shock wave ahead of the cylindrical afterbody, its interaction with the spike boundary layer, leading to flow separation and finally forming a toroidal vortical region near the root of the spike (frame Ⓜ at $\tau + 7\Delta\tau$), as explained by Feszty *et al.*⁴⁰ in his work. A similar phenomenon was also observed in our experimental investigation, as can be seen in frame Ⓜ at $\tau + 9\Delta\tau$. Here the vortical region is numbered 3 (V1) and 4 (V2). Vortical region-V1 comes from the triple point of the lambda- λ shock formed from the spike-tip. On the other hand, vortical region-V2 forms from the triple point of the lambda- λ shock forming closer to the blunt-body surface. The formation of V2 is purely due to the collapse phase, as the compressed gas during collapse accumulates in front of the forebody surface and generates another shock that travels upstream. Feszty *et al.*⁴⁰ believed the vortical region (V2) to be the driving mechanism for pulsation.

Meanwhile, as the CSS advances toward the aft body, the reflected shock emanating from the triple point, as seen in frame Ⓜ at $\tau + 6\Delta\tau$, impinges on the spike body surface and causes the boundary layer to separate, resulting in the formation of a separation bubble and shear layer over it. As the shock system moves downstream, the separation bubble and shear layer grow in size (from Ⓜ at $\tau + 6\Delta\tau$ to frame Ⓜ at $\tau + 10\Delta\tau$). As already discussed, at the triple point, due to velocity gradient, a shear layer exists. Since the pressure behind the bow shock is relatively high compared to the pressure behind the oblique shock wave, the shear layer curls inwards (toward the spike), forming a toroidal vortical region V1, frame Ⓜ at $\tau + 6\Delta\tau$ of Fig. 6. This vortical region is continuously fed from the growing shear layer (S2), resulting in increase in its size, frame Ⓜ to Ⓜ at $\tau + 10\Delta\tau$ of Fig. 6. A significant mass (high density) is trapped inside this vortical region V1. As the shock system moves close to the cylindrical afterbody, the vortical regions V1 and V2 interact with each other and the cylindrical afterbody, resulting in its breakdown and release of the trapped high-density gas within it.

Unlike the earlier findings by Feszty *et al.*,⁴⁰ where it was observed that the gas trapped inside the vertical region V2, causing the rapid expansion, it is clear that it is not just the vortical region V2 near the root of the spike but also the growing vertical region V1, which contributes to the rapid expansion. The interaction of the vortical regions (V1 and V2 and with the wall) followed by the rapid release of the trapped gas inside them results in pressure rise near the root of the spike as seen in Fig. 7(a), marked Ⓜ at $\tau + 11\Delta\tau$. It has to be emphasized here that V1 is formed from the triple point of the λ -shock system arising due to the impingement of the shallow oblique shock on a

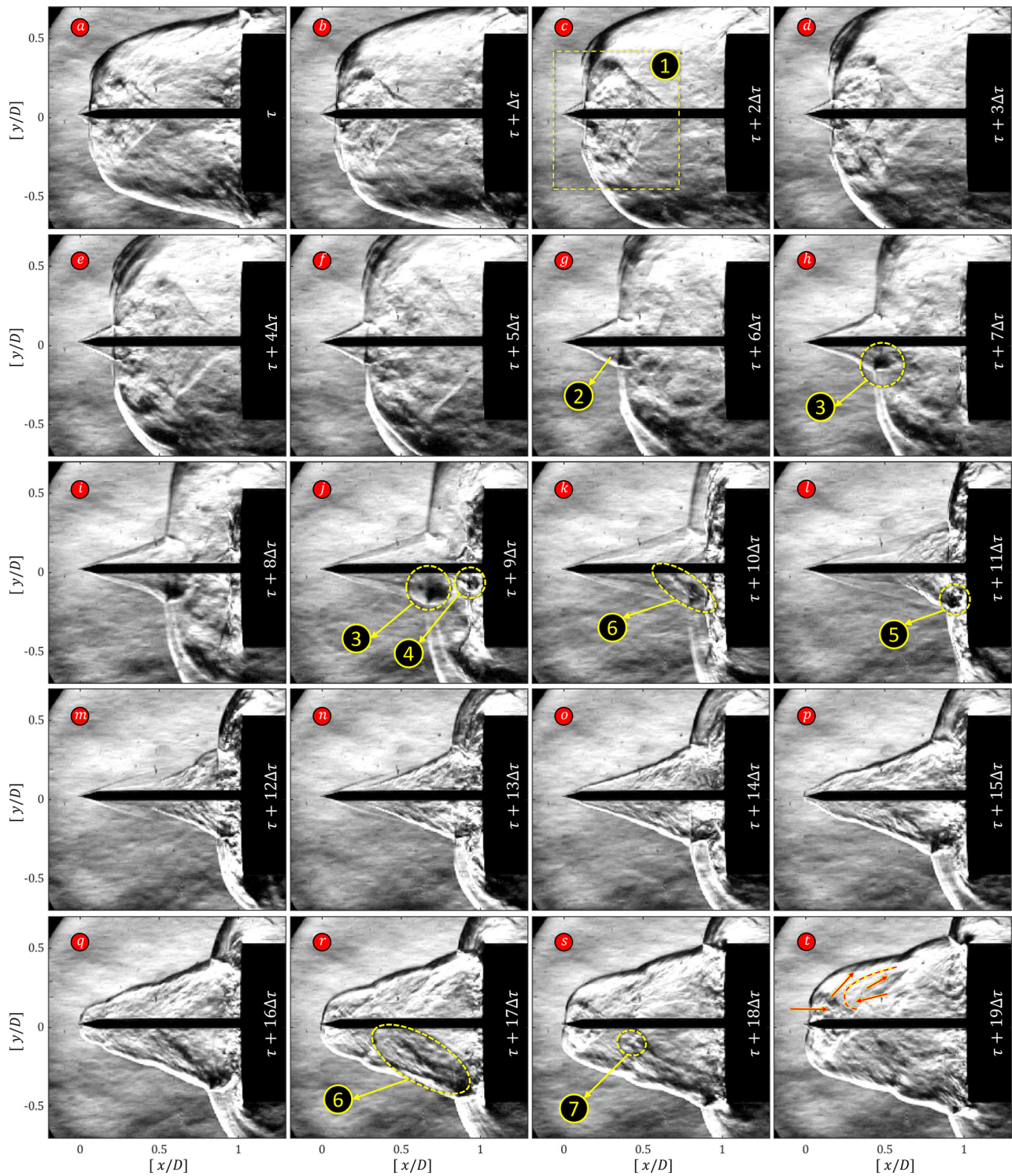


FIG. 6. Instantaneous schlieren image frames representing a pulsating flow cycle from a typical time of τ to $\tau + 19\Delta\tau$ where $\Delta\tau = 25 \mu\text{s}$ for a high Re_D case. Flow is from left to right. Dominant flow features: (1) beetle-leaf-like structure, (2) separation shock, (3) vortical region (V1) from the triple point of λ_1 -shock, (4) vortical region (V2) from the triple point of λ_2 -shock, (5) merging of vortical regions, (6) supersonic jet with shock-cells, and (7) normal shock from the impinging supersonic jet at an oblique angle. Multimedia view: <https://doi.org/10.1063/5.0075583.3>

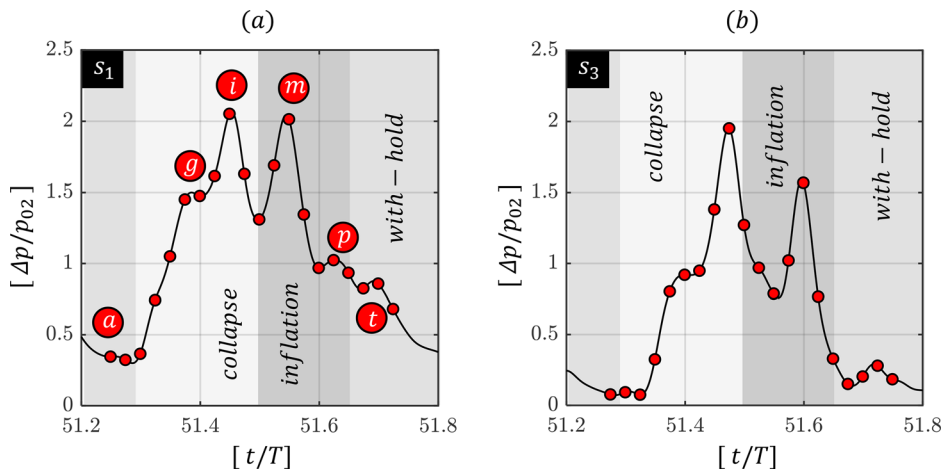


FIG. 7. A typical normalized pressure cycle ($\Delta p/p_{02}$) observed at (a) s_1 and (b) s_3 sensor location during a typical runtime for a high Re_D case. The red color-filled circle and the corresponding text markers represent the event location in accordance with the instantaneous schlieren images available (from τ to $\tau + 19\Delta\tau$) in Fig. 6. The shaded regions demarcate the different regimes of a pulsation cycle.

solid boundary (base body). Similarly, V2 is formed from building another λ -shock system and its corresponding triple point ahead of the base body. A proof of vortices shedding from a λ -shock's triple point is shown in a video available in the [supplementary material](#) for a similar Re_D and $[l/D]$ as evidence through a Detached Eddy Simulation (DES) whose details are beyond the scope of present investigations.

After the release, the trapped gas expands in the lateral and axial direction, starting from frame \textcircled{m} at $\tau + 12\Delta\tau$ till the end of the cycle, marked by a pressure drop on the cylindrical face, as can be seen in Fig. 7(a). As it expands in the lateral direction, the aft body shock approaches close to the cylindrical body, resulting in pressure rise near the shoulder of it as marked \textcircled{p} at $\tau + 12\Delta\tau$ in Fig. 7(b), followed by its drop as the gas expands. The expanding gas toward the upstream direction, named expanding shock system (ESS), is bounded by oblique shock and bow shock wave with a shear layer at their point of interaction, and its first occurrence is seen clearly in frame \textcircled{a} at $\tau + 13\Delta\tau$ of Fig. 6. One should not confuse this shock wave with the oblique shock wave emanating from the spike tip. With the rapid expansion of the gas, this shock system moves in axial (away from aft body) and lateral direction. The shear layer impinges on the cylindrical

afterbody resulting in pressure rise on its surface as can be seen in Fig. 7(a) marked \textcircled{a} at $\tau + 13\Delta\tau$ and in Fig. 7(b) marked \textcircled{a} at $\tau + 17\Delta\tau$. The shock system reaches out to the tip of the spike, where it changes its shape from oblique to bow shock (frame \textcircled{c} at $\tau + 17\Delta\tau$). The ESS remains attached to the tip of the spike, and expansion happens only in the lateral direction, which is the withhold phase of the cycle. The sequence of events mentioned above continues to occur again in the next pulsation cycle.

B. Shock foot-print analysis

The sequence of shock motion and the associated flow physics can be explained through a $x-t$ diagram [see Fig. 8 (Multimedia view)]. A suitable line profile along the flow direction is first drawn to begin constructing the $x-t$ diagram [Fig. 8(a)]. The line is drawn so that the oscillating shock's path or footprint is passing through the line. Such a passing renders the $x-t$ diagram over a while by registering the shock's trace with good contrast [Fig. 8(b)]. The cause mentioned above is the primary reason for not picking a simple straight line parallel to the spike-stem. As described earlier, in Fig. 8(a) the line segment along which the $x-t$ diagram is constructed has been given in dotted

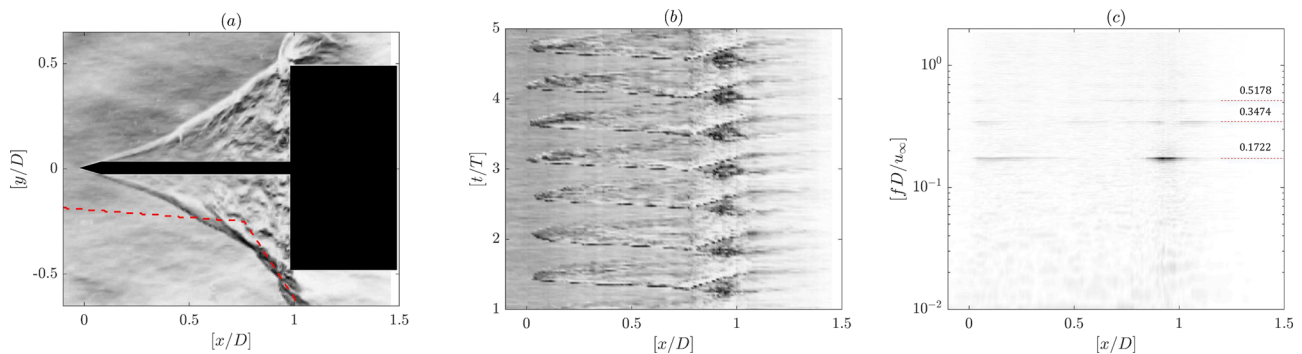


FIG. 8. (a) Extracting the shock foot trajectory by grabbing the image light intensity along the dotted red-line from every schlieren image for the high Re_D case; (b) constructed $x-t$ diagram shown for a typical normalized time scale showing the periodic shock motion ($T = 1$ ms); (c) intensity normalized fast-Fourier transform of the $x-t$ diagram showing the presence of discrete dimensionless frequencies as similar to that of the unsteady pressure signal in Fig. 3(b). Multimedia view: <https://doi.org/10.1063/5.0075583.4>

red-line. The intensity variations along that line in each of the captured images are piled upon to trace the shock motion as shown in Fig. 8(b). A periodic shock oscillation of ten-cycles can be seen in it. By performing a fast Fourier transform (FFT) along the y -axis for all the $[x/D]$ locations, a x - f diagram is constructed, where the dominant frequencies can be observed as shown in Fig. 8(c). The capturing of these frequencies will not be possible if the line segments in Fig. 8(a) are not selected properly. Both visual cues from the high-speed schlieren imaging and the FFT of the unsteady pressure signals thus, help in constructing an appropriate x - t and x - f diagrams. The fundamental frequency $[f_1 D/u_\infty]$ is observed at about 0.172 and two clear overtones are seen at $[f_2 D/u_\infty] \approx 0.3432$ and $f_3 \approx 0.5189$. They exist in a relationship of $[f_3 D/u_\infty] = 3[f_1 D/u_\infty]$ and $[f_2 D/u_\infty] = 2[f_1 D/u_\infty]$, which indicates a self-sustained harmonic behavior of the oscillating shock systems.

In addition to the temporal details about the shock motion, the x - t diagram gives valuable information about the shock velocities. A typical trace of shock trajectory from the x - t diagram given in Fig. 8(b) (fifth cycle) is extracted digitally through an in-house MATLAB program through a combination of the edge-detection algorithm. The three different phases of the pulsation cycle are shown in the extracted trajectory as shown in Fig. 9(a). One of the applicable parameters of this trajectory is that the gradient of it will lead directly to the velocity of shock motion as shown in Fig. 9(b). After the analysis of Fig. 9, it is evident that the shock system accelerates rapidly against the flow direction up to a $[u/U_\infty] \approx 1$ during part of the inflation stage and then decelerates as it nears the spike tip. Later, due to the with-hold stage, the shock stays constant but expands in the transverse direction. Due to it, the velocity values are observed to be zero. At the end of the with-hold stage, the shock system collapses and compresses the recirculation gas against the forebody. At the time of the collapse stage, the shock system accelerates and achieves a maximum velocity of $[u/U_\infty] \approx 0.25$ before the next cycle starts. These analyses stay consistent with the numerical findings of Feszty *et al.*⁴⁰

C. Unsteady pressure signals

In order to understand the underlying flow physics of the pulsation cycle clearly, unsteady pressure signals are acquired at six vital locations [see Fig. 2(b)] symmetric about the axis. A typical pressure cycle observed during pulsation is shown in Fig. 7(a) with all three stages distinctly marked for the high Re_D case. The timestamp of solid red dots represents the time stamp of the instantaneous schlieren snapshots given in Fig. 8 for one-to-one comparison. In Fig. 10, the root mean square (rms) of the pressure variation (pressure loading) is given as solid red-dots, and the length of error-bar represents the pressure fluctuation intensity ($\sqrt{(p - \bar{p})^2} = \sqrt{(\Delta p)^2}$) for two different cases of Re_D . A maximum drop of 98.24% is seen between the high and low Re_D cases in pressure loading and fluctuation intensity. From the time-averaged schlieren image given in Fig. 5(b), the flow looks almost symmetric, and the angle of attack of the model was indeed set to 0° with $\pm 0.1^\circ$. However, the pressure distribution in Fig. 10(b) is not symmetric. One of the primary reasons for this behavior is explained from the point of view of radial shock-related instabilities, which produces rotating stationary waves.⁶⁴ The phenomena mentioned above is a topic by itself, and it will not be considered for further discussion.

A typical trace of pressure history from ten-cycles for each of the sensors (from s_1 to s_6) is given in Figs. 11(a) and 11(c) for both Re_D cases. The asymmetric variations in amplitude and a small phase shift in the signals from the sensors placed symmetrically about the axis can be easily observed due to radial shock-related instabilities as mentioned earlier. The FFT of the signals from the symmetrically placed sensors after spectral averaging ($s_1 + s_4, s_2 + s_5,$ and $s_3 + s_6$) is given in Figs. 11(b) and 11(d) to study the dominant frequencies involved in the pulsation cycle and also to avoid cluttering of similar figures for different Re_D . The fundamental is observed at $[f_1 D/u_\infty] = 0.172$, and the overtones are observed at $[f_2 D/u_\infty] = 0.3403$ and $[f_3 D/u_\infty] = 0.5124$, respectively, exactly in the relation of $[f_3 D/u_\infty] = 3[f_1 D/u_\infty]$ and $[f_2 D/u_\infty] = 2[f_1 D/u_\infty]$ as seen in the x - t diagram

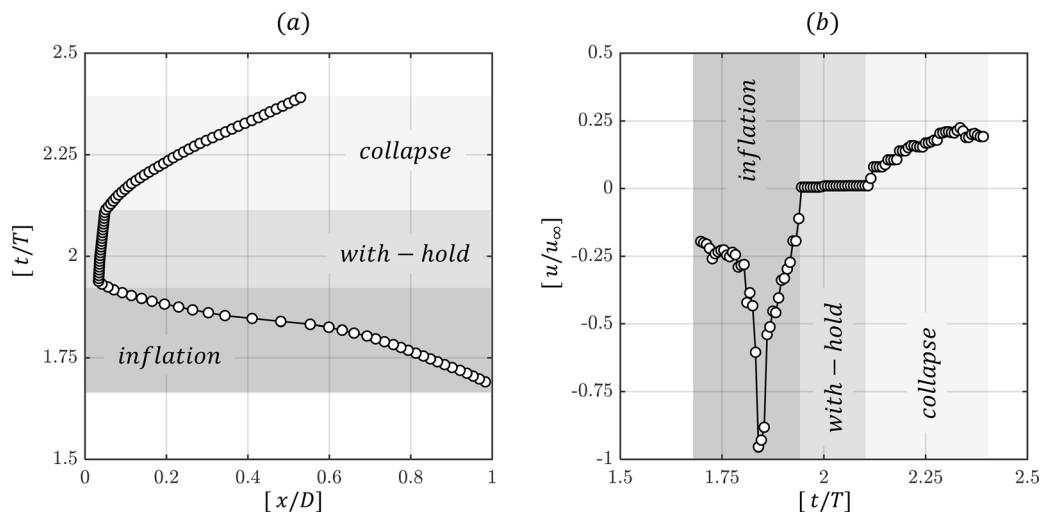


FIG. 9. (a) A typical x - t trace from a cycle of pulsating shock-motion shown in Fig. 8 for a high Re_D case, explaining the presence of distinct three cycles: inflation, with-hold, and collapse; (b) plot showing the variation of shock speed at different time instants during the considered pulsation cycle by measuring the slope along the shock trajectory.

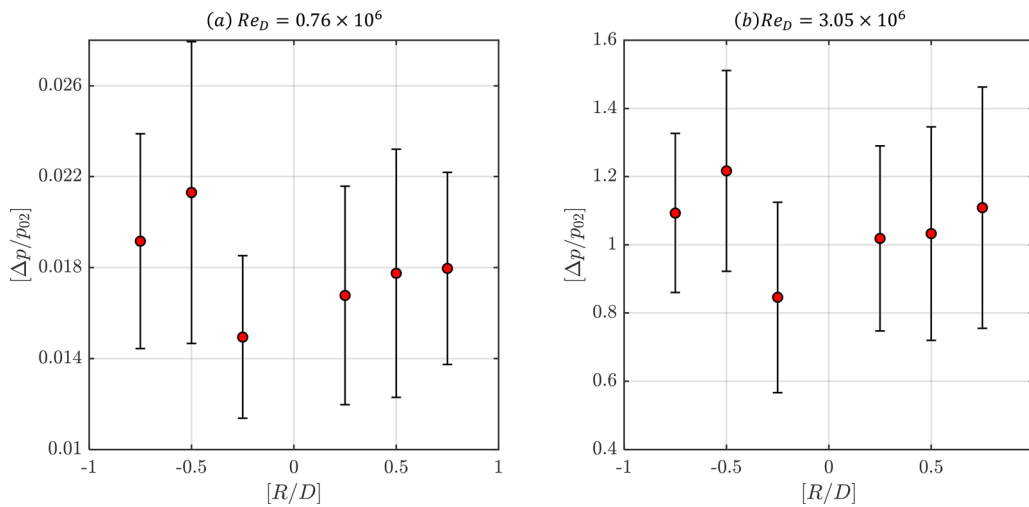


FIG. 10. Normalized pressure measurements on the flat-face spiked body flow showing the rms ($\Delta p_{rms}/p_{02}$, red filled circles) and fluctuation intensity ($\sqrt{\Delta p^2}/p_{02}$, error-bar) for two different cases of Re_D at the same freestream Mach number of $M_\infty = 8.16$: (a) $Re_D = 0.76 \times 10^6$ (low) and (b) 3.01×10^6 (high).

analysis of Fig. 8(c). These similarities will help in corroborating the findings from the schlieren images and unsteady pressure signals. In addition, from the empirical relation given by Kenworthy³⁹ as given below,

$$\left(\frac{f_e D}{u_\infty}\right) = \left[0.25 - 0.067\left(\frac{l}{D}\right)\right], \quad (1)$$

after substituting the values given in Table I, and Fig. 4, the estimated frequency, $[f_e D/u_\infty] = 0.171$, which is almost closer to $[f_1 D/u_\infty]$ from the unsteady pressure measurements by only $\pm 0.6\%$.

Unlike the earlier findings by Feszty *et al.*,⁴⁰ where it was observed that the gas trapped inside the vortical region V2, causing the rapid expansion, it is clear that it is not just the vortical region V2 near the root of the spike but also the growing vertical region V1 contributes to the rapid expansion. The interaction of the vortical regions (V1 and V2 and with the wall) followed by the rapid release of the trapped gas inside them (as seen in Fig. 6 of frame @ at $\tau + 12\Delta\tau$) results in pressure rise near the root of the spike as seen in Fig. 7, where the pressure peaks to $[\Delta p/p_{02}] \sim 2$. The first peak appears at the interaction of the approaching shock and the shock arising due to the compression of collapsing gas against the forebody. The second peak in the pressure cycle arises from the refraction of approaching shock against the forebody wall, which leads to the continuation of the next cycle. The interaction of shock systems sheds interacted vortices back into the recirculation region, which causes rapid inflation. The inflation decreases pressure on the flat-face, as seen in the pressure cycle. At the end of the inflation stage, the pressure curve plateaus for a specific duration and begins to fall further, representing the with-hold stage.

D. Modal analysis

The dynamic events happening during the pulsation cycle can be better understood through the modal analysis.⁶⁵ Two modal decomposition tools are employed to understand the shock motion: (a) proper orthogonal decomposition [POD, $\Phi(x/D, y/D, a_\phi(t/T))$] and (b)

dynamic mode decomposition [DMD, $\Theta(x/D, y/D, t/T)$]. The first method helps understand the energy contents contained in each of the modes and the total number of dominant modes required to represent the flowfield. The second method will help identify the dominant temporal contents and the corresponding spatial modes.

The method involves the preparation of images that will carry only the fluctuations from the fluid phenomena but not the artifacts from the anomalies due to instrumentation. The images are prepared using the procedures mentioned in Karthick *et al.*,⁶⁶ and all the unwanted features like parasite reflections, window defects, and light spots are removed. The processed images are loaded into a column matrix, and 1000 such images are used to construct the complete matrix. Later, through a series of single value and eigenvalue decomposition, POD and DMD modes are extracted by using the procedure thoroughly explained in the book of Kutz *et al.*⁶⁵ Due to the limited intensity fluctuations seen in the high-speed schlieren images of the low Re_D case, the entire set is discarded for modal analysis as no valuable conclusions could be drawn. On the other hand, the high Re_D case is considered given that there are distinct flow features present in the respective high-speed schlieren images.

In Figs. 12(a) and 13(c), the dominant energetic spatial mode [$\Phi_1(x/D, y/D)$] and the FFT of the dominant energetic temporal mode are given. Looking at the spatial mode [Fig. 12(a)], it can be seen that the features represent the time-averaged flow features as seen in Fig. 5(b) of the schlieren image. Thus, it can be concluded that the shock motion in the extremities of the spike tip forms the dominant flow features. The FFT from the temporal mode [Fig. 13(c)] also in agreement with $x-t$ diagram's FFT [Fig. 9(c)] and the FFT of the unsteady pressure signal [Fig. 11(d)].

From the POD analysis, the energy contents in individual modes and the cumulative energy distribution are given in Figs. 13(a) and 13(b). It can be seen that it requires only the first six modes to represent the 25% of flow energy (derived from the fluctuation square of the density gradients). Among them, the first mode [$\Phi_1(x/D, y/D)$]

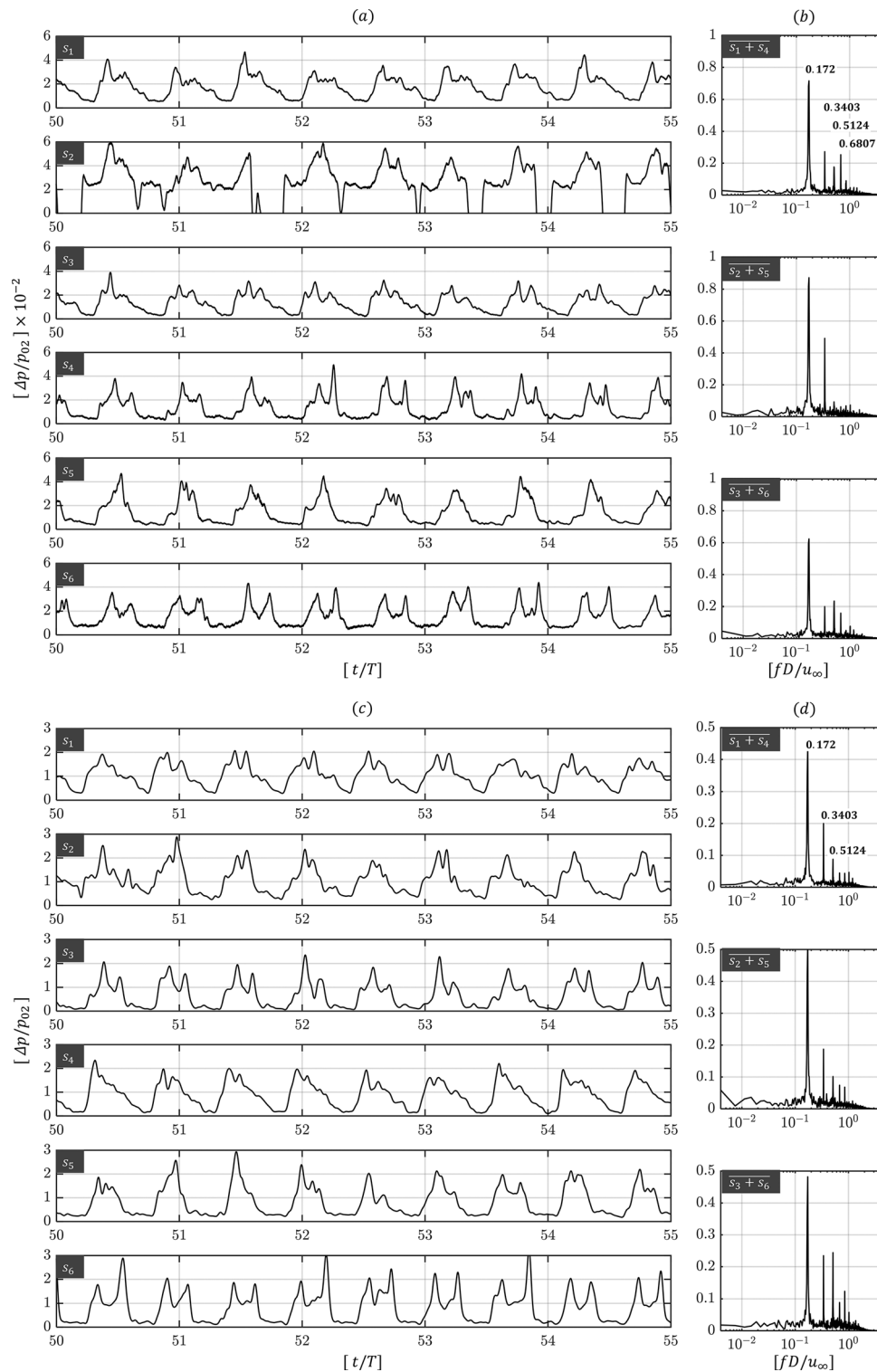


FIG. 11. (a) and (c) Pressure cycles observed at individual locations from s_1 to s_6 [see Fig. 2(b)] during the run-time for two different Re_D cases; (b) and (d) fast-Fourier transform (FFT) of the spectrally averaged pressure signals (considering the symmetry of sensor placement) showing the presence of discrete dimensionless frequencies and normalized pressure amplitudes ($\Delta p / \rho_{02}$) for two different Re_D cases.

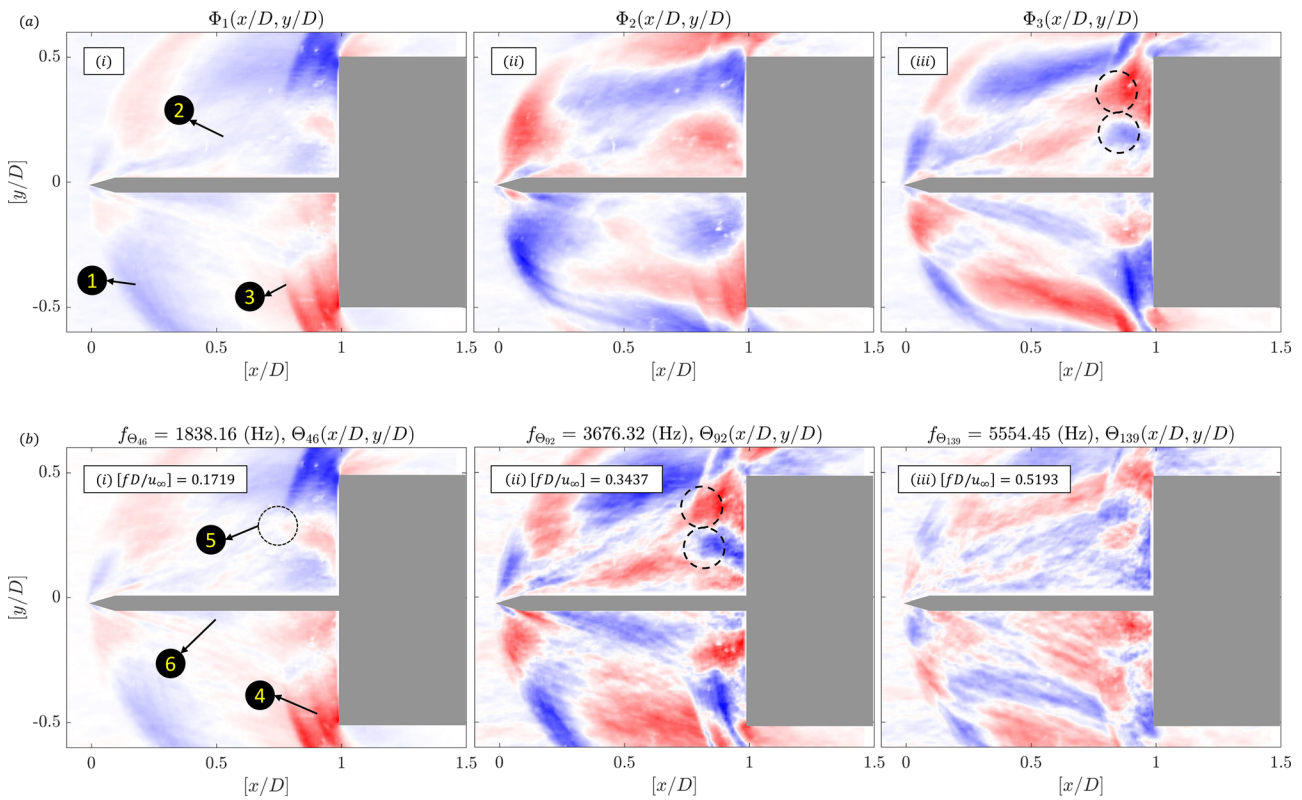


FIG. 12. Contour plots showing the similarity between the (a) first three dominant POD spatial modes $[\Phi_{1-3}(x/D, y/D)]$ and (b) the first three dominant DMD spatial modes: (i) $[f_{\Theta_{46}}D/u_\infty] = 0.1719$, (ii) $[f_{\Theta_{92}}D/u_\infty] = 0.3437$, and (iii) $[f_{\Theta_{139}}D/u_\infty] = 0.5193$. Dominant flow features: (1) inflated shock in the leading edge of the spike; (2) separated free shear layer; (3) collapsed shock forming ahead of the flat-face; (4) shock formation due to the rapid compression of collapsing flowfield; (5) triple point; (6) re-circulation region.

alone contains about 10% of the total energy, whereas the second $[\Phi_2(x/D, y/D)]$ and third $[\Phi_3(x/D, y/D)]$ spatial modes represent 5% and 3% of the total energy, respectively. From analyzing the $\Phi_2(x/D, y/D)$ and $\Phi_3(x/D, y/D)$, the structures represent the inflation phase and the merging of vortices during the collapse phase. The interpretation is drawn by comparing the most distinct spatial features from the energetic spatial modes with that of the instantaneous schlieren images as shown in Fig. 6.

From the DMD analysis, the dominant dynamic spatial mode (12b) provides information on the spatial extent of shock oscillation. The corresponding oscillation frequency is given in the dominant DMD temporal mode 13d. These findings shed similar information as seen from the $x-t$ diagram and the unsteady pressure analysis as before. Since the dominant dynamics are from the shock-motion, the dynamic spatial mode $[\Theta_{46}(x/D, y/D)]$ corresponding to $[f_{46}D/u_\infty] = 0.1719$ as shown in Figs. 12(b-i) represents the same time-averaged flow feature in Fig. 5(b) and the POD dominant energetic spatial mode [see Fig. 12(a-i)]. However, due to the temporal fluctuations, some noise will be observed in the spatial mode. However, for the second dominant temporal mode at $[f_{92}D/u_\infty] = 0.3437$, the formation of distinct vortical regions can be seen in Fig. 13(b-ii) (shown in dotted circles). Similarly, for the third dominant temporal mode $[f_{139}D/u_\infty] = 0.5193$, the corresponding

spatial mode in Fig. 13(b-iii) shows some parts from each division of the cycle, and it is unclear due to large noise.

From the modal analysis, it is clear that the interaction of the V1 and V2 vortical zone drives the shock oscillation. The burst of these structures is observed at a frequency precisely equal to twice the fundamental frequency of the pulsation.

V. CONCLUSIONS

An experimental campaign is carried out to study the pulsating flowfield observed around an axisymmetric flat-face cylinder at zero angles of attack. The experiments are performed in a hypersonic flow generated using the recently converted short-duration hypersonic shock tunnel into a Ludwig tunnel operating for a longer duration. Experiments are done at two different freestream Re_D (0.76 and 3.01×10^6) at a constant M_∞ . High-speed schlieren imaging, unsteady pressure measurements, $x-t$ analysis, spectral analysis, and modal analysis are performed to understand the flow physics. Following are the major conclusions of the present study:

1. From the analysis of high-speed schlieren images at high Re_D , the presence of V1 and V2 vortical zones is identified to drive the collapse stage of the pulsation cycle, which is not observed before experimentally.

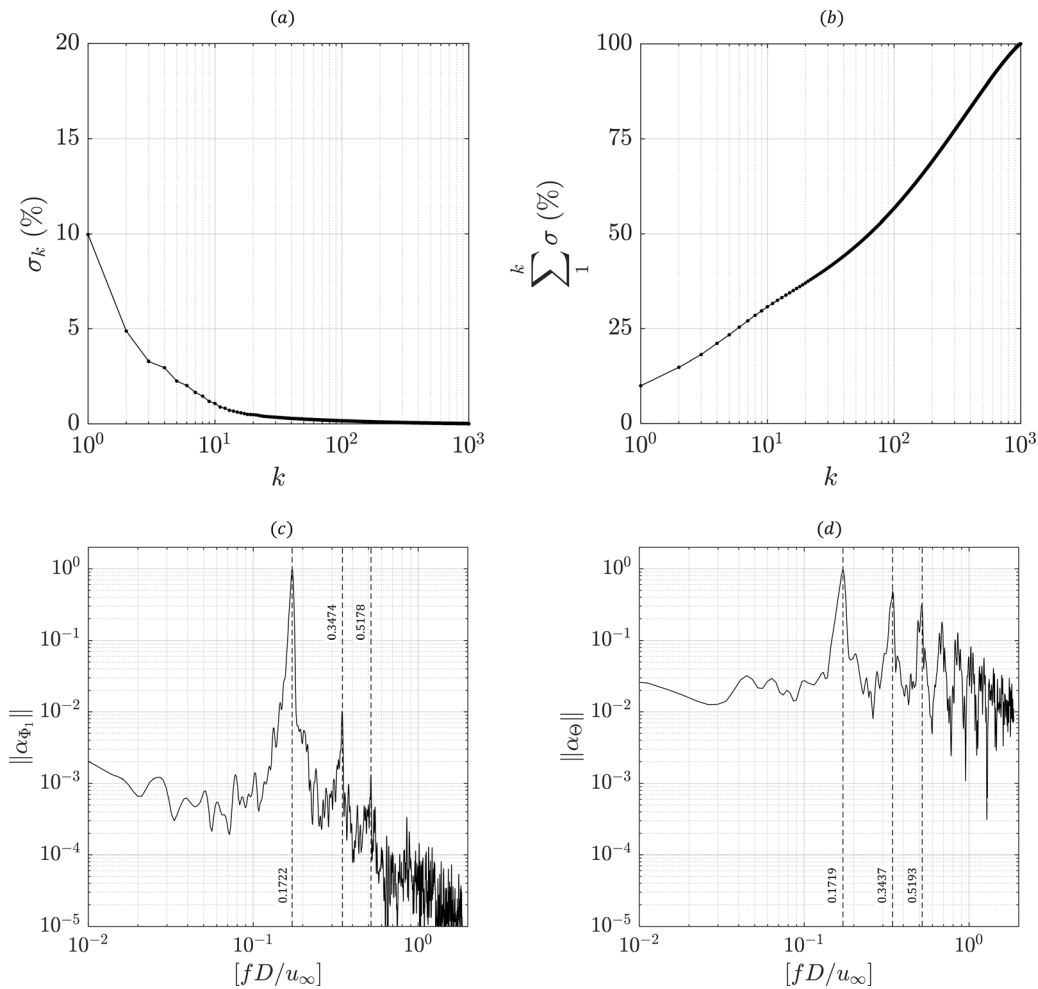


FIG. 13. (a) Plot showing the variation of energy contents (σ_k) in each of the POD $[\Phi(x/D, y/D)]$ modes (k); (b) plot showing the cumulative energy distribution from the POD $[\Phi(x/D, y/D)]$ analysis. (c) Plot showing the presence of discrete frequencies ($[fD/u_\infty]$) at different normalized amplitude ($\|\alpha_{\Phi_1}\|$) from the temporal coefficients variation of the dominant POD temporal mode $[a(\Phi_1(t))]$; (d) plot showing the presence of discrete frequencies ($[fD/u_\infty]$) at different normalized amplitude ($\|\alpha_{\Theta}\|$) from the DMD temporal mode $[\Theta(t/T)]$ analysis.

2. From the $x-t$ diagram, the frequencies of the vortices interactions are identified, and the presence of V1 and V2 vortices is found to be particularly unique to flows with high M_∞ , primarily due to the presence of shallow shock angle and oblique shock impingement on the base body.
3. From the unsteady pressure signal, a maximum drop of 98.24% in pressure loading and fluctuation intensity is seen between the different Re_D cases. Furthermore, the dominant pressure load exists at the time of vortex burst and forebody shock formation.
4. The spectral characteristics observed between the two different Re_D cases reveal invariant dominant frequencies, however, with varying amplitudes and different decay rates. The low and high Re_D cases exhibit an inverse and $-7/3$ decay rate, respectively, at higher frequencies. The observation indicates the formation of possible turbulent structures at high Re_D during vortical breakdowns.

5. From the modal analysis, the third dominant energetic mode and the second dominant dynamic mode comprise the vortical interactions of V1 and V2. The interactions are observed from the DMD analysis at a dimensionless frequency of exactly twice that of the dominant pulsation frequency.

The findings will help prepare the forebody shielding with appropriate materials to avoid acoustic loads or design an efficient active or passive control device to overcome the unsteadiness. The outcome will also be helpful to come up with new geometrical changes to spike or fore-body shape that can modify or diminish these dominant flow patterns and reduce unsteadiness.

SUPPLEMENTARY MATERIAL

See the [supplementary material](#) for the combined vorticity and density contours in the form of a high-speed “video”, and it shows the

shedding of vortical structures from the λ -shock's triple point in a pulsating spiked body at hypersonic flow.

ACKNOWLEDGMENTS

The authors thank the assistance offered by the late assistant Mr. Jeevan of LHSR, IISc-Bengaluru in preparing and instrumenting the models for testing.

AUTHOR DECLARATIONS

Conflict of Interest

The authors report no conflicts of interest.

Authors' Contributions

M.I.S., R.S., S.K.K., and G.J. have contributed equally to this work.

DATA AVAILABILITY

The data that support the findings of this study are available from the corresponding author upon reasonable request.

REFERENCES

- J. Peng, C. Luo, Z. Han, Z. Hu, G. Han, and Z. Jiang, "Parameter-correlation study on shock-shock interaction using a machine learning method," *Aerosp. Sci. Technol.* **107**, 106247 (2020).
- P. K. Seshadri and A. De, "Investigation of shock wave interactions involving stationary and moving wedges," *Phys. Fluids* **32**, 096110 (2020).
- D. Wang, Z. Li, Z. Zhang, N.-S. Liu, J. Yang, and X.-Y. Lu, "Unsteady shock interactions on V-shaped blunt leading edges," *Phys. Fluids* **30**, 116104 (2018).
- A. S. Durna, M. E. H. A. Barada, and B. Celik, "Shock interaction mechanisms on a double wedge at Mach 7," *Phys. Fluids* **28**, 096101 (2016).
- W. Huang, H. Wu, Y. G. Yang, L. Yan, and S. B. Li, "Recent advances in the shock wave/boundary layer interaction and its control in internal and external flows," *Acta Astronaut.* **174**, 103–122 (2020).
- J. Détery and J.-P. Dussauge, "Some physical aspects of shock wave/boundary layer interactions," *Shock Waves* **19**, 453–468 (2009).
- P. M. Ligrani, E. S. McNabb, H. Collopy, M. Anderson, and S. M. Marko, "Recent investigations of shock wave effects and interactions," *Adv. Aerodyn.* **2**, 4 (2020).
- F. Grasso, C. Purpura, B. Chanetz, and J. Détery, "Type III and type IV shock/shock interferences: Theoretical and experimental aspects," *Aerosp. Sci. Technol.* **7**, 93–106 (2003).
- M. L. Mason and S. A. Berry, "Global aeroheating measurements of shock-shock interactions on swept cylinder," *J. Spacecr. Rockets* **53**, 678–692 (2016).
- D. V. Gaitonde, "Progress in shock wave/boundary layer interactions," *Prog. Aerosp. Sci.* **72**, 80–99 (2015).
- D. S. Dolling, "Fifty years of shock-wave/boundary-layer interaction research: What next?," *AIAA J.* **39**, 1517–1531 (2001).
- G. Narayana and S. Selvaraj, "Attenuation of pulsation and oscillation using a disk at mid-section of spiked blunt body," *Phys. Fluids* **32**, 116106 (2020).
- S. Desai, V. P. K. V. Kulkarni, and H. Gadgil, "Universal scaling parameter for a counter jet drag reduction technique in supersonic flows," *Phys. Fluids* **32**, 036105 (2020).
- S. Deep and G. Jagadeesh, "Aerothermodynamic effects of controlled heat release within the hypersonic shock layer around a large angle blunt cone," *Phys. Fluids* **30**, 106103 (2018).
- C. Chen, T. Gao, and J. Liang, "Separation induced low-frequency unsteadiness in a supersonic combustor with single-side expansion," *Phys. Fluids* **31**, 056103 (2019).
- P. Jain and A. Vaidyanathan, "Aero-acoustic feedback mechanisms in supersonic cavity flow with subcavity," *Phys. Fluids* **33**, 126102 (2021).
- N.-H. Liu, X.-R. Li, P.-F. Hao, X.-W. Zhang, and F. He, "Mode switch in tonal under-expanded impinging jets," *Phys. Fluids* **33**, 124102 (2021).
- V. Tekure, P. S. Pophali, and K. Venkatasubbiah, "Numerical investigation of aerospikes semi-cone angle and a small bump on the spike stem in reducing the aerodynamic drag and heating of spiked blunt-body: New correlations for drag and surface temperature," *Phys. Fluids* **33**, 116108 (2021).
- W.-F. Zhou, Y.-C. Hu, M.-Z. Tang, G. Wang, M. Fang, and Y.-G. Yang, "Mechanism of separation hysteresis in curved compression ramp," *Phys. Fluids* **33**, 106108 (2021).
- D. Rockwell and E. Naudascher, "Self-sustained oscillations of impinging free shear layers," *Annu. Rev. Fluid Mech.* **11**, 67–94 (1979).
- S. K. Karthick, "Shock and shear layer interactions in a confined supersonic cavity flow," *Phys. Fluids* **33**, 066102 (2021).
- G. Kumar and A. De, "Modes of unsteadiness in shock wave and separation region interaction in hypersonic flow over a double wedge geometry," *Phys. Fluids* **33**, 036109 (2021).
- W.-Z. Xie, S.-Z. Yang, C. Zeng, K. Liao, R.-H. Ding, L. Zhang, and S. Guo, "Improvement of the free-interaction theory for shock wave/turbulent boundary layer interactions," *Phys. Fluids* **33**, 075104 (2021).
- W. A. Mair, "Experiments on separation of boundary layers on probes in front of blunt-nosed bodies in supersonic air stream," *Philos. Mag.* **43**, 695–716 (1952).
- H.-P. Kabelitz, "Zur Stabilität geschlossener Grenzschichtablösegebiete an konischen Drehkörpern bei Hyperschallanströmung: Mit 5 Tab., 66 Literaturstellen," Ph.D. thesis (Abt. Wiss. Berichtswesen d. Dt. Forschungs-u. Versuchsanst. f. Luft-u, 1971).
- S. M. Bogdonoff and I. E. Vas, "Preliminary investigations of spiked bodies at hypersonic speeds," *J. Aerosp. Sci.* **26**, 65–74 (1959).
- D. Maull, "Hypersonic flow over axially symmetric spiked bodies," *J. Fluid Mech.* **8**, 584–592 (1960).
- M. Y. M. Ahmed and N. Qin, "Recent advances in the aerothermodynamics of spiked hypersonic vehicles," *Prog. Aerosp. Sci.* **47**, 425–449 (2011).
- M. S. Karimi and M. J. Oboodi, "Investigation and recent developments in aerodynamic heating and drag reduction for hypersonic flows," *Heat Mass Transfer/Waerme- Stoffuebertrag.* **55**, 547–569 (2019).
- J. M. Détery, "Shock phenomena in high speed aerodynamics: Still a source of major concern," *Aeronaut. J.* **103**, 19–34 (1999).
- T. F. O'Brien and M. J. Lewis, "Power law shapes for leading-edge blunting with minimal shock standoff," *J. Spacecr. Rockets* **36**, 653–658 (1999).
- R. D. Braun, R. W. Powell, and J. E. Lyne, "Earth aerobraking strategies for manned return from mars," *J. Spacecr. Rockets* **29**, 297–304 (1992).
- B. Venukumar, G. Jagadeesh, and K. P. J. Reddy, "Counterflow drag reduction by supersonic jet for a blunt body in hypersonic flow," *Phys. Fluids* **18**, 118104 (2006).
- V. Kulkarni and K. P. J. Reddy, "Enhancement in counterflow drag reduction by supersonic jet in high enthalpy flows," *Phys. Fluids* **20**, 016103 (2008).
- K. R. Sekar, S. K. Karthick, S. Jegadheeswaran, and R. Kannan, "On the unsteady throttling dynamics and scaling analysis in a typical hypersonic inlet-isolator flow," *Phys. Fluids* **32**, 126104 (2020).
- M. K. K. Devaraj, P. Jutur, S. M. V. Rao, G. Jagadeesh, and G. T. K. Anavardham, "Experimental investigation of unstart dynamics driven by subsonic spillage in a hypersonic scramjet intake at Mach 6," *Phys. Fluids* **32**, 026103 (2020).
- C. Wood, "Hypersonic flow over spiked cones," *J. Fluid Mech.* **12**, 614–624 (1962).
- M. S. Holden, "Experimental studies of separated flows at hypersonic speeds. I—Separated flows over axisymmetric spiked bodies," *AIAA J.* **4**, 591–599 (1966).
- M. Kenworthy, "A study of unstable axisymmetric separation in high speed flows," Ph.D. dissertation (Department of Aerospace and Ocean Engineering, Virginia Polytechnic Institute and State University, 1978).
- D. Feszty, K. J. Badcock, and B. E. Richards, "Driving mechanisms of high-speed unsteady spiked body flows, part 1: Pulsation mode," *AIAA J.* **42**, 95–106 (2004).
- D. Feszty, K. J. Badcock, and B. E. Richards, "Driving Mechanism of High-Speed Unsteady Spiked Body Flows, Part 2: Oscillation Mode," *AIAA J.* **42**(1), 107–113 (2004).
- A. Panaras and D. Drikakis, "High-speed unsteady flows around spiked-blunt bodies," *J. Fluid Mech.* **632**, 69–96 (2009).

- ⁴³A. Antonov, V. Gretsov, and S. Shalaev, "Nonsteady supersonic flow over spiked bodies," *Fluid Dyn.* **11**, 746–751 (1977).
- ⁴⁴D. Sahoo, S. K. Karthick, S. Das, and J. Cohen, "Shock-related unsteadiness of axisymmetric spiked bodies in supersonic flow," *Exp. Fluids* **62**, 89 (2021).
- ⁴⁵L. E. Ericsson and J. P. Reding, *Unsteady Aerodynamic Flow Field Analysis of the Space Shuttle Configuration: Part-III-Unsteady Aerodynamics of Bodies With Concave Nose Geometries* (Lockheed Missiles and Space Company Inc., 1976).
- ⁴⁶J. P. Reding, R. A. Guenther, and B. J. Richter, "Unsteady aerodynamic considerations in the design of a drag reduction spike," *J. Spacecr. Rockets* **14**, 54–60 (1977).
- ⁴⁷X.-R. Geng, Y.-W. Gui, A.-L. Wang, and L.-X. He, "Numerical investigation on drag and heat-transfer reduction using 2-D planar and axisymmetrical forward facing jet," *Kongqi Donglixue Xuebao/Acta Aerodyn. Sin.* **24**, 85–89 (2006).
- ⁴⁸R. C. Mehta, "Numerical analysis of pressure oscillations over axisymmetric spiked blunt bodies at Mach 6.8," *Shock Waves* **11**, 431–440 (2002).
- ⁴⁹R. Mehta, "Numerical simulation of self-sustained oscillations over spiked blunt-bodies," in *39th Aerospace Sciences Meeting and Exhibit* (American Institute of Aeronautics and Astronautics, 2001).
- ⁵⁰V. Karlovskii and V. Sakharov, "Numerical investigation of supersonic flow past blunt bodies with protruding spikes," *Fluid Dyn.* **21**, 437–445 (1986).
- ⁵¹V. Khlebnikov, "Effect of periodic disturbances on the flow in a nonsymmetric separation zone ahead of a blunt body," *Fluid Dyn.* **31**, 731–738 (1996).
- ⁵²B. E. Edney, "Effects of shock impingement on the heat transfer around blunt bodies," *AIAA J.* **6**, 15–21 (1968).
- ⁵³G. S. Settles, *Schlieren and Shadowgraph Techniques: Visualizing Phenomena in Transparent Media*, 1st ed. (Springer-Verlag, Berlin, Heidelberg, 2001).
- ⁵⁴R. Sriram, L. Srinath, M. K. K. Devaraj, and G. Jagadeesh, "On the length scales of hypersonic shock-induced large separation bubbles near leading edges," *J. Fluid Mech.* **806**, 304–355 (2016).
- ⁵⁵K. Oswatitsch, "Der druckrückgewinn bei geschossen mit rückstoßantrieb bei hohen überschallgeschwindigkeiten," *Forsch. Entwicklungen Heerswaffenamates, Bericht 2*, 1005 (1957).
- ⁵⁶P.-J. Lu and L.-T. Jain, "Numerical investigation of inlet buzz flow," *J. Propul. Power* **14**, 90–100 (1998).
- ⁵⁷M. Yamauchi, K. Fujii, and F. Higashino, "Numerical investigation of supersonic flow around a spiked blunt body," *J. Spacecr. Rockets* **32**, 32–42 (1995).
- ⁵⁸H. W. Coleman and W. G. Steele, *Experimentation, Validation, and Uncertainty Analysis for Engineers* (John Wiley & Sons, Inc., 2009).
- ⁵⁹M. A. Sutcliffe and R. G. Morgan, "The measurement of Pitot pressure in high enthalpy expansion tubes," *Meas. Sci. Technol.* **12**, 327–334 (2001).
- ⁶⁰M. D. Santo, C. Liguori, A. Paolillo, and A. Pietrosanto, "Standard uncertainty evaluation in image-based measurements," *Measurement* **36**, 347–358 (2004).
- ⁶¹T. B. Gatski and J.-P. Bonnet, *Compressibility, Turbulence and High Speed Flow* (Academic Press, 2013).
- ⁶²F. Gnani, K. Lo, H. Zare-Behtash, and K. Kontis, "Experimental investigation on shock wave diffraction over sharp and curved splitters," *Acta Astronaut.* **99**, 143–152 (2014).
- ⁶³V. Kariovskii, V. Levin, and V. Sakharov, "Flow past blunt bodies with nose spikes and surface injection," *Fluid Dyn.* **22**, 601–606 (1988).
- ⁶⁴A. Demetriades and A. T. Hopkins, "Asymmetric shock-wave oscillations on spiked bodies of revolution," *J. Spacecr.* **13**(11), 703–704 (1976).
- ⁶⁵J. N. Kutz, S. L. Brunton, B. W. Brunton, and J. L. Proctor, *Dynamic Mode Decomposition: Data-Driven Modeling of Complex Systems* (Society for Industrial and Applied Mathematics, Philadelphia, 2016).
- ⁶⁶S. K. Karthick, S. M. V. Rao, and G. Jagadeesh, "Passive scalar mixing studies to identify the mixing length in a supersonic confined jet," *Exp. Fluids* **58**, 59 (2017).